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GROUP 3200

IN THE UNITED STATES PATENT AND TRADEMARK OFFICE

Applicants: Mannava, et al.)

Serial No: 08/719,341)

Filed: September 25, 1996)

For: LASER SHOCK PEENED GAS TURBINE)
ENGINE COMPRESSOR AIRFOIL EDGES)

Group Art Unit: 3622

Examiner: C. Verdier

20/12
1/15/98

Hon. Assistant Commissioner for Patents
Washington, D.C. 20231

SIR:

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GROUP 3500

AMENDMENT

In response to the Office Action mailed December 2, 1997, please make the below-identified amendments and consider the following remarks:

IN THE SPECIFICATION:

Please amend the specification as follows:

Page 5, line 23; delete "cross-sectional illustrative" and insert --cross-section schematic--.

IN THE CLAIMS:

Please amend the following claims as indicated.

1. (THRICE AMENDED) A gas turbine engine component comprising:

a metallic compressor airfoil having a leading edge and a trailing edge and a pressure side and a suction side,

at least a first laser shock peened surface on a first side of said airfoil,

said laser shock peened surface extending radially along at least a portion of said leading edge and extending chordwise from said leading edge, and

a first region having deep compressive residual stresses

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